Amendments to the Claims:

1.-12. (cancelled).

13. (currently amended) A method for refurbishing a gas turbine blade made from a textured superalloy body coated with a protective coating, the method comprising the steps of:
coating a surface of said body with a high temperature stable surface coating, thereby covering said protective coating;
restoring the microstructure of the superalloy body by performing a solution heat treatment on the body, thereby maintaining said thermally stable surface coating;
removing jointly said surface coating and said protective coating; and
providing a second protective coating on said body.

14. (previously presented) The method according to claim 13,
wherein a γ-phase and a γ'-phase are present in said superalloy and wherein the temperature of said solution heat treatment is at least the solution temperature of the γ' phase.

15. (previously presented) The method according to claim 13,
wherein said solution heat treatment is performed with a temperature above 1100 0C.

16. (currently amended) A method for refurbishing a gas turbine blade made from a textured superalloy body coated with a protective coating, the method comprising the steps of:
removing the protective coating;
coating a surface of said body with a high temperature stable surface coating;
restoring the microstructure of the superalloy body by performing a solution heat treatment on said body, thereby maintaining said thermally stable surface coating;
removing the surface coating; and
providing a second protective coating on said body.

17. (previously presented) The method according to claim 16,
wherein a γ-phase and a γ'-phase are present in the superalloy and wherein the temperature of said solution heat treatment is at least a solution temperature of the γ'-phase.

18. (previously presented) The method according to claim 16,
wherein said solution heat treatment is performed with a temperature above 1100 0C.

19. (previously presented) The method according to claim 13 or 16,
wherein the textured article is a single crystal article.

20. (previously presented) The method according to claim 13 or 16,
wherein the textured article is a directionally solidified article.

21. (previously presented) The method according to claim , 13 or 16,
wherein said surface is applied with an appropriate surface coating.

22. (previously presented) The method according to claim , 13 or 16,
wherein the surface layer is applied to a region which has been newly built up, in particular has been produced by build-up welding.

23. (previously presented) The method according to claim 13 or 16,
wherein the surface layer is applied to a region which surrounds a repaired crack.

24. (previously presented) The method according to claim 13 or 16, wherein a metallic surface layer, in particular of nickel or cobalt is used.

25. (previously presented) The method according to claim 24, wherein the metallic layer is applied by electroplating.

26. (previously presented) The method according to claim 24, wherein the surface layer is applied by cold gas spraying.

27. (currently amended) The method according to claim ~~24,~~ 25 or 26, wherein the surface layer is removed by means of an acid treatment.

28. (previously presented) A method for refurbishing a gas turbine blade made from a textured superalloy body coated with a protective coating, the method comprising the steps of:

coating a surface of said body with a high temperature stable surface coating, thereby covering said protective coating;

performing a solution heat treatment on the body wherein a γ-phase and a γ'-phase are present in said superalloy and wherein the temperature of said solution heat treatment is at least the solution temperature of the γ' phase, thereby maintaining said thermally stable surface coating;

removing jointly said surface coating and said protective coating; and

providing a second protective coating on said body, wherein grain recrystallization is suppressed by providing bulk conditions which assure a higher temperature threshold for grain recrystallization.

29. (previously presented) A method for refurbishing a gas turbine blade made from a textured superalloy body coated with a protective coating, the method comprising the steps of:

removing the protective coating;

coating a surface of said body with a high temperature stable surface coating;

performing a solution heat treatment on said body wherein a γ-phase and a γ‘ phase are present in the superalloy and wherein the temperature of said solution heat treatment is at least a solution temperature of the γ‘-phase, thereby maintaining said thermally stable surface coating;

removing the surface coating; and

providing a second protective coating on said body,

wherein grain recrystallization is suppressed by covering areas with said surface coating.

30. (previously presented) A method for recovering texture of a textured article which is made from a superalloy, comprising the steps of:

creating on a surface of the article a high temperature stable surface coating; and

performing a solution heat treatment on said article wherein a γ-phase and a γ‘-phase are present in said superalloy and wherein the temperature of said solution heat treatment is at least the solution temperature of the γ‘-phase, thereby maintaining said thermally stable surface coating, restoring the microstructure of the textured article, and suppressing grain recrystallization by providing bulk conditions which assure a higher temperature threshold for grain recrystallization.

31. (previously presented) The method according to claim 30, wherein said article is a gas turbine component.

32. (previously presented) The method according to claim 31, wherein said gas turbine component is a blade or a vane.

33. (previously presented) The method according to claim 30, wherein said superalloy is cobalt-based with precipitations or carbides that provide a strengthening mechanism similar to a γ-phase in Nickel based alloys.

34. (new) The method according to claim 24,

wherein the surface layer is removed by means of an acid treatment.